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APPROXIMATE FORMULAS FOR THE COMPUTATION OF

TURBULENT BOUNDARY-LAYER MOMENTUM

THICKNESSES IN COMPRESSIBLE FLOWS

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APPROXIMATE FORMULAS FOR THE COMPUTATION OF

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SUMMARY

Approximate formulas for the computation of the momentum thicknesses of turbulent boundary layers on two-dimensional bodies, on bodies of revolution at zero angle of attack, and on the inner surfaces of round channels all in compressible flow are given in the form of integrals that can be conveniently computed. The formulas involve the assumptions that the momentum thickness may be computed by use of a boundary-layer velocity profile which is fixed and that skin-friction formulas for flat plates may be used in the computation of boundary-layer thicknesses in flow with pressure gradients. The effect of density changes on the ratio of the displacement thickness to the momentum thickness of the boundary layer is taken into account. Use is made of the experimental finding that the skin-friction coefficient for turbulent flow is independent of Mach number. The computations indicate that the effect of density changes on the momentum thickness in flows with pressure gradients is small for subsonic flows.

INTRODUCTION

A number of methods are available for the computation of boundary-layer momentum thicknesses for incompressible flow. The increasing importance of flows at Mach numbers approaching and exceeding 1 has emphasized the need of formulas that would make possible the comparatively rapid computation of boundary-layer momentum thicknesses for compressible flows. The purpose of the present work is therefore to provide approximate formulas for the computation of boundary-layer-thickness parameters for

compressible flows. The present work furnishes no new
information concerning the boundary-layer shape, skin-
friction coefficient, position of the transition point,
or likelihood of boundary-layer separation.

Approximate formulas for the computation of the
momentum thicknesses of turbulent boundary layers on
two-dimensional bodies, on bodies of revolution at zero
angle of attack, and on the inner surfaces of round
channels all in compressible flow are given in the form
of integrals that can be conveniently computed. The
approximate formulas contain the assumptions that the
momentum thickness may be computed by use of a boundary-
layer velocity profile which is fixed during the inte-
gration and that skin-friction formulas for flat plates
may be used in the computation of boundary-layer momentum
thicknesses for flow with pressure gradients. The
formulas are applicable to all unseparated, turbulent
boundary layers and in special cases to laminar boundary
layers. The numerical values of the ratio of the dis-
placement thickness to the momentum thickness, a ratio
that appears in the momentum equation and that is capable
of specifying approximately the velocity distribution
through the turbulent boundary layer in incompressible
flow, are corrected for density changes in the boundary
layer by use of low-speed velocity distributions. Use is
made of the experimental finding that the skin-friction
coefficient in turbulent flow is independent of Mach
number.

The problem of computing the boundary-layer momentum
thickness for compressible flow has been treated by Young
and Winterbottom (reference 1), who integrated the
boundary-layer momentum equation for laminar flow by using
the skin-friction relation from the Pohlhausen theory
(reference 2, p. 109), fixing the velocity profile, and
correcting the density through the boundary layer for the
effects of compressibility. For the turbulent boundary
layer, the momentum equation was integrated by a step-by-
step process in which a fixed velocity profile was used
and the effect of density changes through the boundary
layer on the ratio of the displacement thickness to the
momentum thickness was ignored.

The problem of computing the momentum thickness over
a body of revolution for incompressible flow has been
treated by Young (reference 3), who computed the momentum
thickness of the laminar boundary layer by a step-by-step

computation in which an extension of the Pohlhausen method
was used. The thickness of the turbulent part of the
boundary layer was computed by a step-by-step process in
which a fixed velocity profile and the momentum equation
for a body of revolution were used.

In order to substantiate the assumption that skin-
friction formulas for turbulent flow along flat plates
may be used in the computation of momentum thicknesses
for flow with pressure gradients, references 3 to 6 are
cited. In these references, good agreement between calcu-
lated and experimental results was generally obtained
although fairly large adverse pressure gradients were
present in many of the cases.

The assumption that the momentum thickness may be
computed to a close approximation by fixing the velocity
profile during integration is substantiated by the work
in references 3 and 4 and by that in reference 6, which
contains a comparison between the computed and experi-
mental values of the momentum thickness over the entire
chord.

That the skin-friction coefficient for turbulent flow
is independent of Mach number is established by the work
in references 7 and 8. Frössel (reference 7) presents
experimental data for turbulent flow in pipes which show
that the velocity profiles for subsonic compressible flow
and the skin-friction coefficients for subsonic and super-
sonic compressible flow do not differ noticeably from
those for incompressible flow. Theodorsen and Regier
(reference 8), by experimenting with rotating disks, showed
that the skin-friction coefficient for turbulent boundary
layers is independent of Mach number. Keenan and Neuman
(reference 9), after performing experiments with pipes,
reached conclusions that did not contradict those of
references 7 and 8.

SYMBOLS

a   constant in equation relating H_c to λ and H_i

b   slope of velocity distribution

c_d drag coefficient per unit span

c_O velocity of sound in free stream

c_p specific heat at constant pressure, foot-pounds per pound-mass per degree

H ratio of displacement thickness to momentum thickness (δ^*/θ)

K constant

k constant in skin-friction formula

L length of airfoil, body of revolution, or round channel; measured along chord or axis of revolution

M_O free-stream Mach number

m exponent in formula for boundary-layer velocity distribution

m' particular value of m

n exponent in skin-friction formula

p static pressure

R gas constant

R_L Reynolds number $(U_O L/v_O)$

R_x Reynolds number based on length of plate $(U_O x/v_O)$

R_θ Reynolds number based on momentum thickness $(U\theta/v)$

r radial distance of point from axis of body of revolution or round channel

r_t radius of body of revolution or round channel

$r_{t_{max}}$ maximum radius of body of revolution or round channel

T absolute temperature

T_O absolute temperature of free stream

T_δ absolute temperature at edge of boundary layer

U   velocity parallel to x at outer edge of boundary layer

U_0 free-stream velocity

U_x value of U at station at which value of $\left(\dfrac{\theta}{L}\right)_x$ is obtained

U_{x_0} value of U at $x = 0$

U_λ value of U chosen to make value of $\left(\dfrac{U_\lambda}{U_x}\right)^{\Delta H_1} - 1$ a maximum

U_1 value of U at x_0

u   velocity inside boundary layer and parallel to surface

w   exponent in formula for viscosity

x   distance measured along surface from forward stagnation point

x_0 position on surface at beginning of integration

y   distance measured normal to x

α   angle between tangent to surface of body of revolution or round channel and axis of revolution

$\beta = r_t\theta + \Omega \cos \alpha$

β_1 value of β at x_0

γ   ratio of specific heat at constant pressure to specific heat at constant volume

δ   nominal thickness of boundary layer

δ^* displacement thickness $\left[\displaystyle\int_0^\delta \left(1 - \frac{\rho u}{\rho_\delta U}\right) dy\right]$

θ   momentum thickness $\left[\displaystyle\int_0^\delta \frac{\rho u}{\rho_\delta U}\left(1 - \frac{u}{U}\right) dy\right]$

θ_1 value of θ at x_0

$$\lambda = \sqrt{\frac{\emptyset}{(U/U_O)^2}}$$

μ coefficient of viscosity

μ_O free-stream viscosity

υ kinematic viscosity at outer edge of boundary
            layer

υ_O kinematic viscosity in free stream

ρ density

ρ_O free-stream density

ρ_δ density at edge of boundary layer

τ_O surface shearing stress

$$\emptyset = 1 + \frac{2}{(\gamma - 1)M_O^2}$$

$$\psi = r_t\theta - \Omega \cos \alpha$$

ψ_1 value of ψ at x_0

$$\Omega = \int_0^\delta \frac{\rho u}{\rho_\delta U}\left(1 - \frac{u}{U}\right)y \, dy$$

$$\Omega^* = \int_0^\delta \left(1 - \frac{\rho u}{\rho_\delta U}\right)y \, dy$$

Subscripts:

c   compressible flow

i   incompressible flow

ANALYSIS

Momentum equation for compressible flow about a two-dimensional body.- The boundary-layer momentum equation for two-dimensional compressible flow (reference 10, p.132)

is given, when the static-pressure variation across the
boundary layer is negligible, as

$$\frac{\partial}{\partial x} \int_0^\delta \rho u^2 \, dy - U \frac{\partial}{\partial x} \int_0^\delta \rho u \, dy = - \tau_0 - \delta \frac{\partial p}{\partial x} \qquad (1)$$

From the equation of motion for compressible, inviscid
flow the relation between the velocity and pressure
derivatives outside the boundary layer may be written
for convenience as

$$\frac{\partial \rho_\delta U^2}{\partial x} - U \frac{\partial \rho_\delta U}{\partial x} = - \frac{\partial p}{\partial x}$$

Then by use of the equation for $\frac{\partial p}{\partial x}$, the definition for
the momentum thickness

$$\theta = \int_0^\delta \frac{\rho u}{\rho_\delta U} \left(1 - \frac{u}{U}\right) \, dy$$

and the definition for the displacement thickness

$$\delta^* = \int_0^\delta \left(1 - \frac{\rho u}{\rho_\delta U}\right) \, dy$$

equation (1) can be written in the form given in reference 1
as

$$\frac{\partial \theta}{\partial x} + \theta \left(\frac{H_c + 2}{U} \frac{\partial U}{\partial x} + \frac{1}{\rho_\delta} \frac{\partial \rho_\delta}{\partial x}\right) = \frac{\tau_0}{\rho_\delta U^2} \qquad (2)$$

where $H_c = \dfrac{\delta^*}{\theta}$.

The principles used in the derivation of equation (2)
were the conservation of mass and Newton's law of motion.
Equation (1) is therefore applicable to both subsonic and
supersonic flow. This equation is not, however, to be
used for flow through a shock wave because in this case
the assumptions of the boundary-layer theory may not be
applicable.

Momentum equation for compressible flow about a body
of revolution.- The boundary-layer momentum equation for
compressible flow about a body of revolution (reference 10,
p. 133) can be written, when the static-pressure variation
across the boundary layer is negligible, as

$$\frac{\partial}{\partial x} \int_0^\delta \rho u^2 r \, dy - U \frac{\partial}{\partial x} \int_0^\delta \rho u r \, dy = - \tau_0 r_t - \frac{\partial p}{\partial x} \int_0^\delta r \, dy \quad (3)$$

Since

$$r = r_t + y \cos \alpha$$

where r_t and $\cos \alpha$ are dependent on x only (fig. 1),
equation (3) may be rewritten as

$$\left(\frac{\partial}{\partial x} \int_0^\delta \rho u^2 r_t \, dy - U \frac{\partial}{\partial x} \int_0^\delta \rho u r_t \, dy \right)$$

$$+ \left[\frac{\partial}{\partial x} \int_0^\delta \rho u^2 (\cos \alpha) y \, dy - U \frac{\partial}{\partial x} \int_0^\delta \rho u (\cos \alpha) y \, dy \right] =$$

$$- \frac{\partial p}{\partial x} \int_0^\delta r_t \, dy - \frac{\partial p}{\partial x} \int_0^\delta (\cos \alpha) y \, dy - r_t \tau_0$$

If θ and Ω are defined by the relations

$$r_t \theta \rho_\delta U^2 = \int_0^\delta \rho u (U - u) r_t \, dy$$

and

$$\Omega (\cos \alpha) \rho_\delta U^2 = \int_0^\delta \rho u (U - u) (\cos \alpha) y \, dy$$

and the equation of motion for compressible, inviscid flow

$$\frac{\partial p}{\partial x} = - \rho_\delta U \frac{\partial U}{\partial x}$$

is used, then the momentum equation becomes

$$- \frac{\partial U}{\partial x} \left(\int_0^\delta \rho_\delta U r_t \ dy - \int_0^\delta \rho u r_t \ dy \right) - \frac{\partial}{\partial x} \left(r_t \theta \rho_\delta U^2 \right)$$

$$- \frac{\partial U}{\partial x} \left[\int_0^\delta \rho_\delta U (\cos \alpha) y \ dy - \int_0^\delta \rho u (\cos \alpha) y \ dy \right]$$

$$- \frac{\partial}{\partial x} \left[\Omega (\cos \alpha) \rho_\delta U^2 \right] = - r_t \tau_o$$

Then δ^* and Ω^* may be defined by the relations

$$r_t \delta^* \rho_\delta U = \int_0^\delta \left(\rho_\delta U - \rho u \right) r_t \ dy$$

and

$$\Omega^* (\cos \alpha) \rho_\delta U = \int_0^\delta \left(\rho_\delta U - \rho u \right) (\cos \alpha) y \ dy$$

so that the momentum equation becomes

$$\frac{\partial}{\partial x} \left(r_t \theta + \Omega \cos \alpha \right) + \frac{1}{U} \frac{\partial U}{\partial x} \left[r_t \theta \left(H_c + 2 \right) + \Omega (\cos \alpha) \left(\frac{\Omega^*}{\Omega} + 2 \right) \right]$$

$$+ \frac{1}{\rho_\delta} \frac{\partial \rho_\delta}{\partial x} \left(r_t \theta + \Omega \cos \alpha \right) = \frac{r_t \tau_o}{\rho_\delta U^2} \qquad (4)$$

In order to permit integration of the momentum equation for compressible flow about a body of revolution, equation (4) should be written in the same form as that for two-dimensional compressible flow (equation (2)). The approximation is therefore made that for flow on the body of revolution

$$\left[r_t \theta (H_c + 2) + \Omega (\cos \alpha) \left(\frac{\Omega^*}{\Omega} + 2 \right) \right] = \left(H_c + 2 \right) (r_t \theta + \Omega \cos \alpha) K$$

or

$$\frac{1 + \dfrac{\Omega \cos \alpha}{r_t \theta}\left(\dfrac{\dfrac{\Omega^*}{\Omega} + 2}{H_c + 2}\right)}{1 + \dfrac{\Omega \cos \alpha}{r_t \theta}} = K$$

In order to determine the magnitude of the ratio $\dfrac{\dfrac{\Omega^*}{\Omega} + 2}{H_c + 2}$, the assumption is made that the velocity distributions through the boundary layer may be approximated by power curves of the type

$$\frac{u}{U} = \left(\frac{y}{\delta}\right)^{1/m}$$

When the flow is incompressible, the definitions of Ω^*, Ω, and H_i may be used to obtain

$$\frac{\Omega^*}{\Omega} = \frac{1 + m}{m}$$

and

$$H_i = \frac{2 + m}{m}$$

Therefore

$$\frac{\dfrac{\Omega^*_i}{\Omega_i} + 2}{H_i + 2} = \frac{3m + 1}{3m + 2}$$

for incompressible flow.

From

$$\int_0^1 F\left(y^{2/p}\right) y \, dy = \frac{1}{2} \int_0^1 F\left(y^{1/p}\right) \, dy$$

it can be shown that

$$\left(\frac{\Omega^*}{\Omega}\right)_{m=m'} = H_{m=2m'}$$

for compressible as well as for incompressible flow.
By use of

$$\left(\frac{\Omega^*}{\Omega}\right)_{m=m'} = H_{m=2m'}$$

and figure 2, which is discussed later in connection with
the effect of compressibility on H_c, the ratio $\dfrac{\frac{\Omega^*}{\Omega} + 2}{H_c + 2}$.
can be evaluated for compressible flow. For all cases of
unseparated flow, the value of the ratio $\dfrac{\frac{\Omega^*}{\Omega} + 2}{H_c + 2}$ differs
from unity by less than 20 percent and, for most cases, by
roughly 10 percent. Thus, even when $\dfrac{\Omega \cos \alpha}{r_t \theta}$ is not a
small fraction, the approximation that $K = 1$ is not
far from true. When, in addition, it is noted that

$$\frac{\Omega \cos \alpha}{r_t \theta} < \frac{\delta}{r_t}$$

and is therefore small over most of the body, the approxi-
mation that $K = 1$ is permissible. If $K = 1$ is used
in equation (4), it becomes

$$\frac{\partial}{\partial x}\left(r_t \theta + \Omega \cos \alpha\right) + \frac{H_c + 2}{U}\frac{\partial U}{\partial x}\left(r_t \theta + \Omega \cos \alpha\right) + \frac{1}{\rho_\delta}\frac{\partial \rho_\delta}{\partial x}\left(r_t \theta + \Omega \cos \alpha\right) =$$

$$\frac{r_t \tau_0}{\rho_\delta U^2}$$

or by substitution of β for $r_t \theta + \Omega \cos \alpha$

$$\frac{\partial \beta}{\partial x} + \beta\left(\frac{H_c + 2}{U}\frac{\partial U}{\partial x} + \frac{1}{\rho_\delta}\frac{\partial \rho_\delta}{\partial x}\right) = \frac{r_t \tau_0}{\rho_\delta U^2} \tag{5}$$

This equation for incompressible flow has been given in
reference 3.

The principles used in the derivation of equation (5)
were the conservation of mass and Newton's law of motion.
The approximation was made that for flow on the body $K = 1$.

Momentum equation for compressible flow over inner
surface of a round channel.- When the flow in a round
channel is such that the flow throughout the boundary
layer is approximately parallel to the wall and a region
exists outside the boundary layer in which viscosity has
no effect, the boundary-layer momentum equation for this
flow is the same as equation (3). In the momentum equa-
tion for flow over the inner surface of a round channel,
the pressure change across the boundary layer is assumed
to be negligible. All of the symbols in the momentum
equation for compressible flow over the inner surface of
a round channel have their meanings unchanged except for
the distance y, which is positive when measured inward
from the wall so that

$$r = r_t - y \cos \alpha$$

The derivation of the momentum equation for flow over
the inner surface of a round channel follows the same pro-
cedure and involves the same assumptions as the derivation
of equation (5). The derived equation is

$$\frac{\partial \psi}{\partial x} + \psi \left(\frac{H_c + 2}{U} \frac{\partial U}{\partial x} + \frac{1}{\rho_\delta} \frac{\partial \rho_\delta}{\partial x} \right) = \frac{r_t \tau_0}{\rho_\delta U^2} \qquad (6)$$

where

$$\psi = r_t \theta - \Omega \cos \alpha$$

Effect of compressibility on H_c.- Since the ratio
of the displacement thickness to the momentum thickness
H_c occurs in the momentum equation, the effect of com-
pressibility upon this ratio should be considered. The
equation for H_c is

$$H_c = \frac{\delta^*}{\theta}$$

$$= \frac{\int_0^\delta \left(1 - \frac{\rho u}{\rho_\delta U} \right) dy}{\int_0^\delta \frac{\rho u}{\rho_\delta U} \left(1 - \frac{u}{U} \right) dy}$$

The density variation through the boundary layer can
be obtained from the velocity distribution through the
boundary layer by restricting the treatment to the case
in which there is no heat flow through the surface and
the effective Prandtl number is equal to unity. The
equation

$$c_p T + \frac{u^2}{2} = \text{constant}$$

is then applicable to flow in the boundary layer. When
the additional restriction is made that the static-pressure
variation through the boundary layer is negligible, the
expression for $\frac{\rho}{\rho_\delta}$ becomes

$$\frac{\rho}{\rho_\delta} = \frac{1 + \left[1 - \left(\frac{U}{U_o}\right)^2\right]\frac{\gamma - 1}{2}M_o^2}{1 + \left[1 - \left(\frac{u}{U}\right)^2\left(\frac{U}{U_o}\right)^2\right]\frac{\gamma - 1}{2}M_o^2}$$

and the expression for θ becomes

$$\frac{\theta}{\delta} = \int_0^1 \frac{1 + \left[1 - \left(\frac{U}{U_o}\right)^2\right]\frac{\gamma - 1}{2}M_o^2}{1 + \left[1 - \left(\frac{u}{U}\right)^2\left(\frac{U}{U_o}\right)^2\right]\frac{\gamma - 1}{2}M_o^2}\frac{u}{U}\left(1 - \frac{u}{U}\right)d\frac{y}{\delta}$$

which can be reduced to

$$\frac{\theta}{\delta} = \left[\frac{1 + \frac{\gamma - 1}{2}M_o^2}{\left(\frac{U}{U_o}\right)^2\frac{\gamma - 1}{2}M_o^2} - 1\right]\int_0^1 \frac{\frac{u}{U}\left(1 - \frac{u}{U}\right)d\frac{y}{\delta}}{\frac{1 + \frac{\gamma - 1}{2}M_o^2}{\left(\frac{U}{U_o}\right)^2\frac{\gamma - 1}{2}M_o^2} - \left(\frac{u}{U}\right)^2}$$

Let

$$\frac{1 + \frac{\gamma - 1}{2} M_0^2}{\left(\frac{U}{U_0}\right)^2 \frac{\gamma - 1}{2} M_0^2} = \lambda^2$$

or

$$\lambda^2 = \frac{\phi}{\left(\frac{U}{U_0}\right)^2}$$

where

$$\phi = 1 + \frac{2}{(\gamma - 1) M_0^2}$$

then

$$\frac{\theta}{\delta} = \left(\lambda^2 - 1\right) \int_0^1 \frac{\frac{u}{U}\left(1 - \frac{u}{U}\right)}{\lambda^2 - \left(\frac{u}{U}\right)^2} \, d\frac{y}{\delta} \tag{7}$$

By application of the same procedure to the definition of δ^*/δ, then

$$\frac{\delta^*}{\delta} = 1 - (\lambda^2 - 1) \int_0^1 \frac{\frac{u}{U}}{\lambda^2 - \left(\frac{u}{U}\right)^2} \, d\frac{y}{\delta} \tag{8}$$

Large values of λ^2 mean small values of M_0.

The assumption was made that

$$\frac{u}{U} = \left(\frac{y}{\delta}\right)^{1/m}$$

and values were chosen for λ^2; δ^*/δ and θ/δ were then calculated for a range of λ between 1.5 and 14.0 with $m = 3, 4, 5,$ and 7. The curves of $1/H_c$ are given in figure 2. Since

$$H_i = \frac{2 + m}{m}$$

and since H appears in the momentum equation, the curves
of figure 2 are designated by values of H_i rather than m.
Power curves are used merely for convenience in computing
the effect of compressibility on H.

The value of H_c for the Blasius flat-plate profile
for laminar boundary layers (reference 2, p. 88) has been
computed for various values of λ and a Prandtl number
of unity by use of equations (7) and (8). The computations
were repeated with the velocity distribution for laminar
flow over a flat plate at $M_O = 2$ (reference 11). When
H_c was plotted against λ, the results of both computa-
tions were practically identical for small values of λ.
Only the results obtained for the Blasius profile are
therefore presented in figure 3.

Equations (7) and (8) show that although the velocity
distribution through the boundary layer is assumed to be
independent of position along the surface, H_c may vary
with surface position because of its dependence on λ.
For integration of the momentum equations, the dependence
of H_c on λ may be taken into account by approximating
the curves of figures 2 and 3 by the equation

$$H_c = \frac{a}{\lambda - 1} + H_i$$

$$= \frac{a}{\left(\dfrac{\sqrt{\phi}}{\dfrac{U}{U_O}}\right) - 1} + H_i \qquad (9)$$

The values of a chosen to fit the curves of figures 2
and 3 with sufficient accuracy over most of the range of
λ are plotted against H_i in figure 4.

Integration of momentum equation for two-dimensional
flow.- Before equation (2) can be integrated, $(H_c + 2)$,
$\dfrac{1}{\rho_\delta} \dfrac{\partial \rho_\delta}{\partial x}$, and $\dfrac{\tau_O}{\rho_\delta U^2}$ should be replaced by functions of the
velocity distribution over the body, the free-stream Mach
number, and the momentum thickness. The term $H_c + 2$ is
replaced by its equivalent $\dfrac{a}{\left(\dfrac{\sqrt{\phi}}{\dfrac{U}{U_O}}\right) - 1} + (H_i + 2)$. By use

of the gas law

$$\rho_\delta = \frac{p}{RT_\delta}$$

the equation of motion

$$\frac{\partial p}{\partial x} = - \rho_\delta U \frac{\partial U}{\partial x}$$

the Bernoulli equation for compressible flow

$$c_p T_\delta + \frac{U^2}{2} = c_p T_0 + \frac{U_0^2}{2}$$

and the equation for the velocity of sound

$$c_0^2 = (\gamma - 1) c_p T_0$$

the term $\dfrac{1}{\rho_\delta} \dfrac{\partial \rho_\delta}{\partial x}$ can be written as

$$\frac{1}{\rho_\delta} \frac{\partial \rho_\delta}{\partial x} = \frac{-2\left(\frac{U}{U_0}\right) \frac{\partial \left(\frac{U}{U_0}\right)}{\partial x}}{(\gamma - 1) \left[\phi - \left(\frac{U}{U_0}\right)^2 \right]} \tag{10}$$

The local skin-friction coefficient $\dfrac{\tau_0}{\rho_\delta U^2}$ is expressed as a power function of the local Reynolds number based on the boundary-layer momentum thickness by

$$\frac{\tau_0}{\rho_\delta U^2} = \frac{k}{R_\theta^{\,n}} = k \left(\frac{v}{U}\right)^n \theta^{-n} \tag{11}$$

By analogy with the work of reference 11, the viscosity and density used to calculate R_θ are those at the outer edge of the boundary layer. The viscosity at the outer edge of the boundary layer is assumed to be given by

$$\frac{\mu}{\mu_0} = \left(\frac{T_\delta}{T_0}\right)^w \tag{12}$$

where $w = 0.768$ for air (reference 11).

The density at the outer edge of the boundary layer is given by

$$\frac{\rho_0}{\rho_\delta} = \left[\frac{\phi - \left(\frac{U}{U_0}\right)^2}{\phi - 1}\right]^{\frac{1}{1-\gamma}}$$

(13)

in which the flow outside the boundary layer is assumed to be adiabatic. Equations (12) and (13) are used to give the kinematic viscosity v in equation (11) as a function of the velocity distribution over the body and the free-stream Mach number. Equation (2) may then be written as

$$\frac{\partial \theta}{\partial x} + \theta \left\{ \frac{a}{\sqrt{\phi} - \frac{U}{U_0}} \frac{\partial \frac{U}{U_0}}{\partial x} + \frac{H_i + 2}{\left(\frac{U}{U_0}\right)} \frac{\partial \left(\frac{U}{U_0}\right)}{\partial x} - \frac{2\left(\frac{U}{U_0}\right) \frac{\partial \left(\frac{U}{U_0}\right)}{\partial x}}{(\gamma - 1)\left[\phi - \left(\frac{U}{U_0}\right)^2\right]} \right\} = \frac{kv^n}{\left(\frac{U}{U_0}\right)^n \theta^n U_0^n}$$

This equation is a differential equation of the Bernoulli type. When it is made linear and integrated by standard methods, the result is

$$\left(\frac{\theta}{L}\right)_{x/L} = \frac{\left[\sqrt{\phi} - \left(\frac{U}{U_0}\right)_{x/L}\right]^a}{\left[\left(\frac{U}{U_0}\right)_{x/L}\right]^{(H_i+2)} \left[\phi - \left(\frac{U}{U_0}\right)_{x/L}^2\right]^{\frac{1}{\gamma-1}}} \left\{ \frac{\left[\frac{k(1+n)}{R_L^n}\right]}{(\phi-1)^{n\left(w-\frac{1}{\gamma-1}\right)}} \right.$$

$$\int_{x_0/L}^{x/L} \frac{\left(\frac{U}{U_0}\right)^{\left[(H_i+1)(1+n)+1\right]} \left[\phi - \left(\frac{U}{U_0}\right)^2\right]^{\left(nw+\frac{1}{\gamma-1}\right)}}{\left(\sqrt{\phi} - \frac{U}{U_0}\right)^{a(1+n)}} d\frac{x}{L}$$

$$+ \left. \frac{\left(\frac{\theta_1}{L}\right)^{1+n} \left(\frac{U_1}{U_0}\right)^{(H_i+2)(1+n)} \left[\phi - \left(\frac{U_1}{U_0}\right)^2\right]^{\frac{1+n}{\gamma-1}}}{\left[\sqrt{\phi} - \frac{U_1}{U_0}\right]^{a(1+n)}} \right\}^{\frac{1}{1+n}}$$

(14)

where θ_1/L and U_1/U_0 are the values of θ/L and U/U_0 at $\frac{x}{L} = \frac{x_0}{L}$. The value of ϕ is always greater than 1. By use of the Bernoulli equation for compressible flow, $\left(\frac{U}{U_0}\right)^2$ can be shown to be always less than ϕ. When M_0 approaches zero, equation (14) becomes equation (1) of reference 6, that is, the integrated momentum equation for incompressible flow.

Integration of momentum equation for flow over a body of revolution.- The assumptions concerning H_0 are the same as those for the two-dimensional case. The equations for $\frac{\mu}{\mu_0}$, $\frac{\rho_0}{\rho_\delta}$, and $\frac{1}{\rho_\delta}\frac{\partial \rho_\delta}{\partial x}$ are also the same as for the two-dimensional case. The expression for the skin friction, however, involves the approximation that θ can be replaced by $\frac{\beta}{r_t}$. This approximation is equivalent to neglecting the term $\frac{\Omega}{r_t}\cos\alpha$ in the equation

$$\theta = \frac{\beta}{r_t} - \frac{\Omega}{r_t}\cos\alpha$$

Since the value of $\frac{\Omega}{r_t}\cos\alpha$ is always less than $\frac{\delta}{r_t}\theta$, it follows that the approximation is justified only in cases in which $\frac{\delta}{r_t}$ is a small fraction of unity. In regions near the tail, therefore, the accuracy of the approximation for the skin friction may be expected to decrease. The momentum equation for flow over a body of revolution (equation (5)) indicates, however, that the contribution of the skin friction to the boundary-layer thickness becomes less important as the tail is approached. The approximation that θ can be replaced by $\frac{\beta}{r_t}$ is therefore allowable and the term $\frac{r_t \tau_0}{\rho_\delta U^2}$ in equation (5) may be written

$$\frac{r_t \tau_0}{\rho_\delta U^2} = \frac{k r_t^{n+1} \upsilon^n}{\beta^n U^n} \tag{15}$$

When the terms H_c, $\dfrac{1}{\rho_\delta}\dfrac{\partial \rho_\delta}{\partial x}$, and $\dfrac{r_t \tau_0}{\rho_\delta U^2}$ are replaced by equations (9), (10), and (15), then equation (5) becomes

$$\frac{\partial \beta}{\partial x} + \beta \left\{ \frac{a}{\sqrt{\emptyset} - \dfrac{U}{U_0}} \frac{\partial \dfrac{U}{U_0}}{\partial x} + \frac{H_i + 2}{\left(\dfrac{U}{U_0}\right)} \frac{\partial \dfrac{U}{U_0}}{\partial x} - \frac{2\left(\dfrac{U}{U_0}\right)\partial \dfrac{\overline{U}}{U_0}}{(\gamma - 1)\left[\emptyset - \left(\dfrac{U}{U_0}\right)^2\right]} \right\} =$$

$$\frac{r_t^{1+n} k v^n}{\left(\dfrac{U}{U_0}\right)^n U_0^n \epsilon^n}$$

This equation is a differential equation of the Bernoulli type. When it is made linear and integrated by standard methods, the result is

$$\left(\frac{\beta}{r_{t_{max}}^2}\right)_{x/L} = \frac{\left[\sqrt{\emptyset} - \left(\dfrac{U}{U_0}\right)_{x/L}\right]^a}{\left[\left(\dfrac{U}{U_0}\right)_{x/L}\right]^{(H_i+2)}\left[\emptyset - \left(\dfrac{U}{U_0}\right)_{x/L}^2\right]^{\frac{1}{\gamma-1}}} \left\{ \frac{\dfrac{k(1+n)}{R_L^{\,n}}\left(\dfrac{L}{r_{t_{max}}}\right)^{1+n}}{(\emptyset - 1)^n\left(w - \dfrac{1}{\gamma-1}\right)} \right.$$

$$\int_{x_0/L}^{x/L} \frac{\left(\dfrac{U}{U_0}\right)^{\left[(H_i+1)(1+n)+1\right]}\left[\emptyset - \left(\dfrac{U}{U_0}\right)^2\right]^{\left(nw+\frac{1}{\gamma-1}\right)}\left(\dfrac{r_t}{r_{t_{max}}}\right)^{1+n}}{\left[\sqrt{\emptyset} - \left(\dfrac{U}{U_0}\right)\right]^{a(1+n)}} \, d\frac{x}{L}$$

$$\left. + \frac{\left(\dfrac{\beta_1}{r_{t_{max}}^2}\right)^{(1+n)}\left(\dfrac{U_1}{U_0}\right)^{(H_i+2)(1+n)}\left[\emptyset - \left(\dfrac{U_1}{U_0}\right)^2\right]^{\frac{1+n}{\gamma-1}}}{\left[\sqrt{\emptyset} - \left(\dfrac{U_1}{U_0}\right)\right]^{a(1+n)}} \right\}^{\frac{1}{1+n}} \qquad (16)$$

where $\dfrac{\beta_1}{r_{t_{max}}^2}$ and U_1/U_0 are the values of $\dfrac{\beta}{r_{t_{max}}^2}$ and U/U_0 at $\dfrac{x}{L} = \dfrac{x_0}{L}$.

Integration of momentum equation for flow over inner surface of a round channel.- The equation resulting from the integration of equation (6), when the same procedure and assumptions made to integrate equation (5) are used, may be obtained from equation (16) by replacing β by ψ. In the equation for ψ, the quantities denoting reference conditions, U_0, rt_{max}, and M_0, are the conditions at a convenient reference section of the channel and the length L is any convenient length.

The definite integrals occurring in equations (14) and (16) and the equation for ψ may be evaluated by either an analytical or a graphical method, whichever is more convenient.

DISCUSSION

Before equation (14), equation (16), or the integrated equation containing ψ can be used, it is necessary to know the velocity distribution over the surface, the constants in the skin-friction formula, and the value to choose for H_i. The velocity distribution must be that for the Mach number and Reynolds number for which the computation is being made.

Skin-friction formulas.- For the turbulent boundary layer, a power function of R_θ is used for the skin-friction formula. Because references 7 and 8 indicate no noticeable effect of Mach number on skin friction, a skin-friction formula for low speeds may be chosen and approximated by

$$\frac{\tau_0}{\rho_\delta U^2} = \frac{k}{R_\theta^{\,n}}$$

as outlined in reference 6. One formula of the required form is that of Falkner (reference 12)

$$\frac{\tau_0}{\rho_\delta U^2} = \frac{0.006535}{R_\theta^{1/6}}$$

If the skin-friction data available are given in the form of $c_d \sim R_x$, they may be converted to $\dfrac{\tau_0}{\rho_\delta U^2} \sim R_\theta$ by use

of the relations

$$R_\theta = \frac{c_d}{2} R_x$$

and

$$\frac{\tau_o}{\rho_\delta U^2} = \frac{1}{2} \left(c_d + R_x \frac{dc_d}{dR_x} \right)$$

Laminar boundary layer.- Although equation (14),
equation (16), and the equation for ψ have been derived
for turbulent boundary layers, these equations may be used
for the approximate computation of θ, β, and ψ in
laminar boundary layers by the appropriate choice of a
value for H_i and of a skin-friction formula. Although
by use of the skin-friction relation for the Blasius flat-
plate profile the effect of pressure gradient on the skin-
friction coefficient is neglected, the error introduced
is small so long as the average pressure gradient over the
extent of the laminar boundary layer is small. The skin-
friction relation then is

$$\frac{\tau_o}{\rho_\delta U^2} = \frac{0.220}{R_\theta}$$

where $k = 0.220$ and $n = 1$. Although a small decrease
occurs in k as the Mach number increases (reference 11),
the value of k for incompressible flow (0.220) may be
used in view of the approximation already made concerning
the skin friction.

Choice of H_i.- In references 1, 3, and 4, the value
of the boundary-layer shape parameter H was restricted
to 1.4. If equation (14) is used, H_i is given an
arbitrary increment ΔH_i, and $\frac{\theta_1}{L} = 0$, then the change
in $\left(\frac{\theta}{L} \right)_x$ may be given by

$$\frac{\Delta \left(\frac{\theta}{L} \right)_x}{\left(\frac{\theta}{L} \right)_x} = \left(\frac{U_\lambda}{U_x} \right)^{\Delta H_i} - 1$$

where U_λ lies between the maximum and minimum velocity
in the interval x, x_0. The term $\left(\frac{U_\lambda}{U_x} \right)^{\Delta H_i}$ has been

evaluated for incompressible flow and a straight-line
velocity distribution

$$U = U_{x_0} \pm bx$$

The expression for $\left(\dfrac{U_\Delta}{U_x}\right)^{\Delta H_i}$ becomes

$$\left(\frac{U_\Delta}{U_x}\right)^{\Delta H_i} = \left\{ \left[\frac{1 - \left(\dfrac{U_{x_0}}{U_x}\right)^{\left[(H_i+1)(n+1)+2+\Delta H_i(n+1)\right]}}{1 - \left(\dfrac{U_{x_0}}{U_x}\right)^{\left[(H_i+1)(n+1)+2\right]}} \right] \left[\frac{(H_i+1)(n+1)+2}{(H_i+1)(n+1)+2+\Delta H_i(n+1)} \right] \right\}^{\frac{1}{n+1}} \quad (17)$$

Although equation (17) is for incompressible flow and the
expression for $\dfrac{\Delta(\theta/L)_x}{\theta_i(\theta/L)_x}$ is only approximately correct for

cases in which $\dfrac{\theta}{L} \neq 0$, equation (17) indicates that
excessive errors in θ/L are possible if a in equation (9)
is made to equal zero and H_i is replaced by H_c and restricted
to a constant low-speed value when the actual variation of
H_c is large. Increasing the body thickness, the lift
coefficient, or the Mach number increases the variation of
H_c over the surface.

      In the integrated equations the value of H_c may
vary over the surface but that of H_i is assumed to be
constant. Because H_i can usually be estimated to within
0.2 for cases in which flow separation does not occur, the
error in θ/L that may be caused by a poor choice of H_i
is unlikely to be more than 10 percent.

      When the velocity increases in the direction of flow,
the value of H_i may be taken as 1.2. When the velocity
decreases in the direction of flow, the value of H_i
should be increased from 1.4, for cases in which the total
change of velocity is about 20 percent of the initial
velocity, to values near 1.7, for cases in which the total
change in velocity exceeds 30 percent of the initial
velocity. Because the velocity profile for the laminar

boundary layer has been restricted to the Blasius flat-plate profile, H_i is equal to 2.6. For the laminar boundary layer, a then has the value 1.2.

Full thickness of boundary layer.- The full thickness of the turbulent boundary layer in the case of two-dimensional flow may be obtained by use of the relation

$$\frac{\delta}{L} = \left(\frac{\theta}{L}\right)\left(\frac{\delta}{\theta}\right)$$

For the body of revolution, the full thickness of the turbulent boundary layer may be obtained, after $\dfrac{\beta}{r_{t_{max}}^2}$ is computed, by use of the relation

$$\frac{\delta}{r_{t_{max}}} = \frac{-\left(\frac{r_t}{r_{t_{max}}}\right)\left(\frac{\theta}{\delta}\right) + \sqrt{\left(\frac{r_t}{r_{t_{max}}}\right)^2\left(\frac{\theta}{\delta}\right)^2 + 4\left(\frac{\beta}{r_{t_{max}}^2}\right)\left(\frac{\Omega}{\delta^2}\right)\cos\alpha}}{2\left(\frac{\Omega}{\delta^2}\right)\cos\alpha} \tag{18}$$

Curves of θ/δ and Ω/δ^2 against λ are given in figures 5 and 6 for four values of H_i.

The variation of θ/δ for the Blasius flat-plate profile is given in figure 7 for use in computing the full thickness of the laminar boundary layer. For flow about a body of revolution, Ω/δ^2 may be neglected since the laminar boundary layers are usually thin. The expression for the full thickness of the laminar boundary layer on a body of revolution therefore becomes

$$\frac{\varepsilon}{r_{t_{max}}} = \left(\frac{\beta}{r_{t_{max}}^2}\right)\left(\frac{r_{t_{max}}}{r_t}\right)\left(\frac{\delta}{\theta}\right) \tag{19}$$

For flow over the inner surface of a round channel when the boundary layer is thin, equations (18) and (19) are replaced by

$$\frac{\delta}{r_{t_{max}}} = \frac{\left(\frac{r_t}{r_{t_{max}}}\right)\left(\frac{\theta}{\delta}\right) - \sqrt{\left(\frac{r_t}{r_{t_{max}}}\right)^2\left(\frac{\theta}{\delta}\right)^2 - 4\left(\frac{\psi}{r_{t_{max}}^2}\right)\left(\frac{\Omega}{\delta^2}\right)\cos\alpha}}{2\left(\frac{\Omega}{\delta^2}\right)\cos\alpha} \tag{20}$$

and

$$\frac{\delta}{r_{t_{max}}} = \left(\frac{\psi}{r_{t_{max}}^2}\right)\left(\frac{r_{t_{max}}}{r_t}\right)\left(\frac{\delta}{\theta}\right) \tag{21}$$

In order to obtain a general statement regarding the effect of density changes on the momentum thickness, computations of θ were made for a linear velocity distribution and a range of Mach number. The velocity distribution for $0 \leqq \frac{x}{L} \leqq 1$ was defined by

$$\frac{U}{U_O} = 1 + b\frac{x}{L}$$

The only variable was M_O. The curves of θ against M_O are given in figure 8 for $b = 0$ and ± 0.2. The results indicate that the effect of density variation becomes important only at Mach numbers exceeding unity.

CONCLUSIONS

A comparatively rapid method is presented for the computation of boundary-layer momentum thicknesses for flow over two-dimensional bodies, over bodies of revolution at zero angle of attack, and over the inner surface of round channels all in compressible flow.

The computations indicate that the effect of density changes on the momentum thickness in flows with pressure gradients is small for subsonic flows.

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REFERENCES

1. Young, A. D., and Winterbottom, N. E.: Note on the
   Effect of Compressibility on the Profile Drag of
   Aerofoils in the Absence of Shock Waves. Rep.
   No. B. A. 1595, R.A.E. (British), May 1940.

2. Prandtl, L.: The Mechanics of Viscous Fluids. Vol. III
   of Aerodynamic Theory, div. G, secs. 14 and 17,
   W. F. Durand, ed., Julius Springer (Berlin), 1935,
   pp. 84-90 and 102-112.

3. Young, A. D. : The Calculation of the Total and Skin
   Friction Drags of Bodies of Revolution at Zero
   Incidence. R. & M. No. 1874, British A.R.C., 1939.

4. Squire, H. B., and Young, A. D.: The Calculation of
   the Profile Drag of Aerofoils. R. & M. No. 1838,
   British A.R.C., 1938.

5. von Doenhoff, Albert E., and Tetervin, Neal: Determi-
   nation of General Relations for the Behavior of
   Turbulent Boundary Layers. NACA ACR No. 3G13, 1943.

6. Tetervin, Neal: A Method for the Rapid Estimation of
   Turbulent Boundary-Layer Thicknesses for Calculating
   Profile Drag. NACA ACR No. L4G14, 1944.

7. Frössel, W.: Flow in Smooth Straight Pipes at
   Velocities above and below Sound Velocity. NACA TM
   No. 844, 1938.

8. Theodorsen, Theodore, and Regier, Arthur: Experiments
   on Drag of Revolving Disks, Cylinders, and Stream-
   line Rods at High Speeds. NACA ACR No. L4F16, 1944.

9. Keenan, Joseph H., and Neumann, Ernest P.: Friction
   in Pipes at Supersonic and Subsonic Velocities.
   NACA TN No. 963, 1945.

10. Fluid Motion Panel of the Aeronautical Research
    Committee and Others: Modern Developments in Fluid
    Dynamics. Vol. I., S. Goldstein, ed., The Clarendon
    Press (Oxford), 1938.

11. Brainerd, J. G., and Emmons, H. W.: Effect of Vari-
    able Viscosity on Boundary Layers, with a Dis-
    cussion of Drag Measurements. Jour. Appl. Mech.,
    vol. 9, no. 1, March 1942, pp. A-1 - A-6.

12. Falkner, V. M.: A New Law for Calculating Drag. The
    Resistance of a Smooth Flat Plate with Turbulent
    Boundary Layer. Aircraft Engineering, vol. XV,
    no. 169, March 1943, pp. 65-69.

Fig. 1
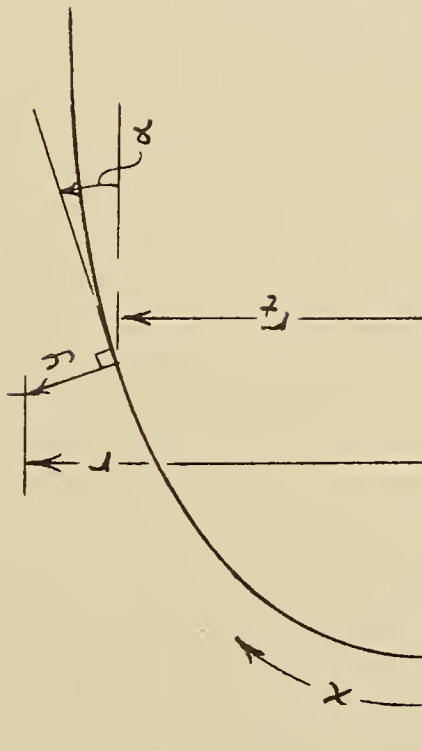
Figure 1.- Coordinates on body of revolution.

Fig. 2   NACA ACR No. L6A22
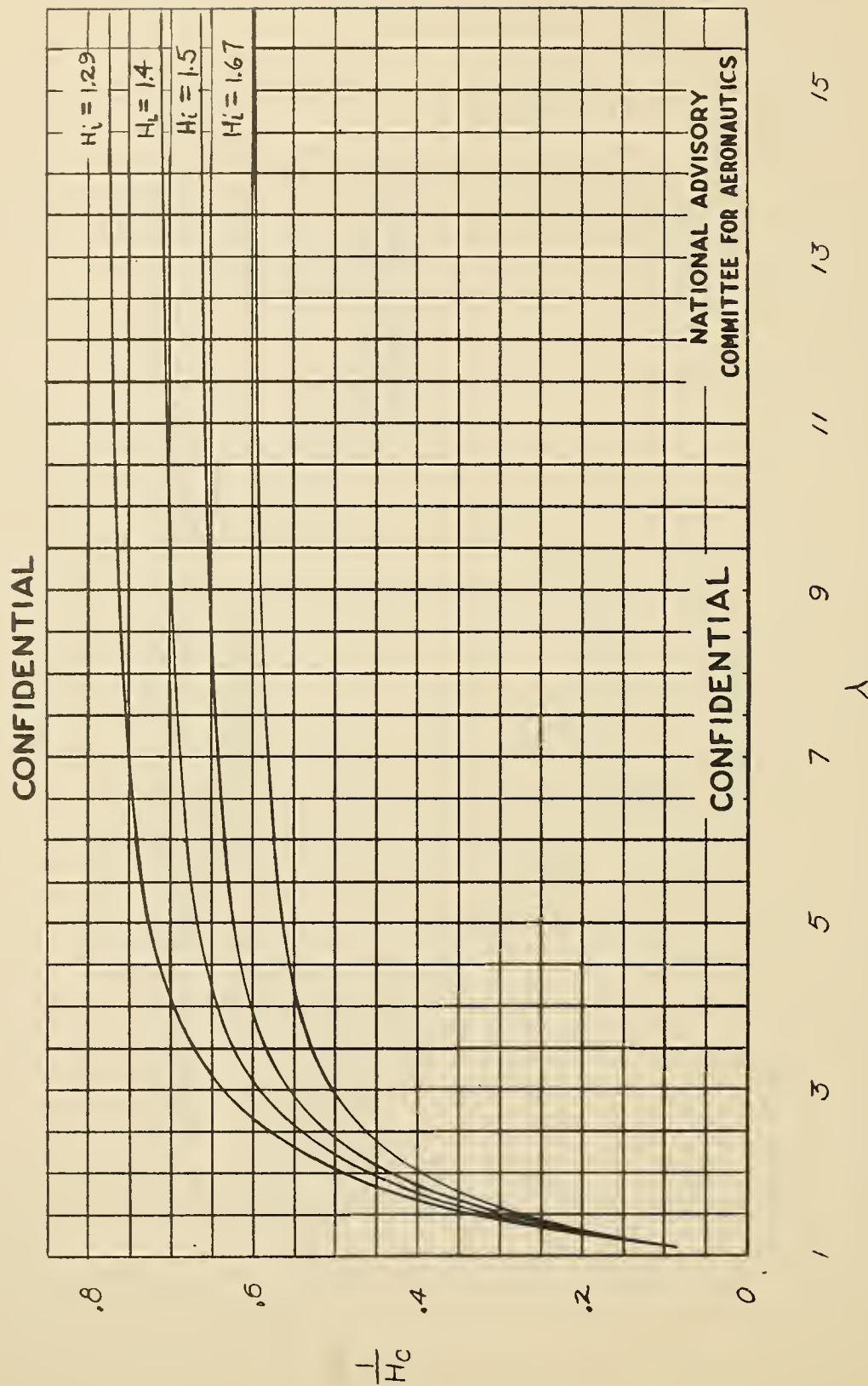
Figure 2.- Variation of $1/H_c$ with λ for various values of H_1.
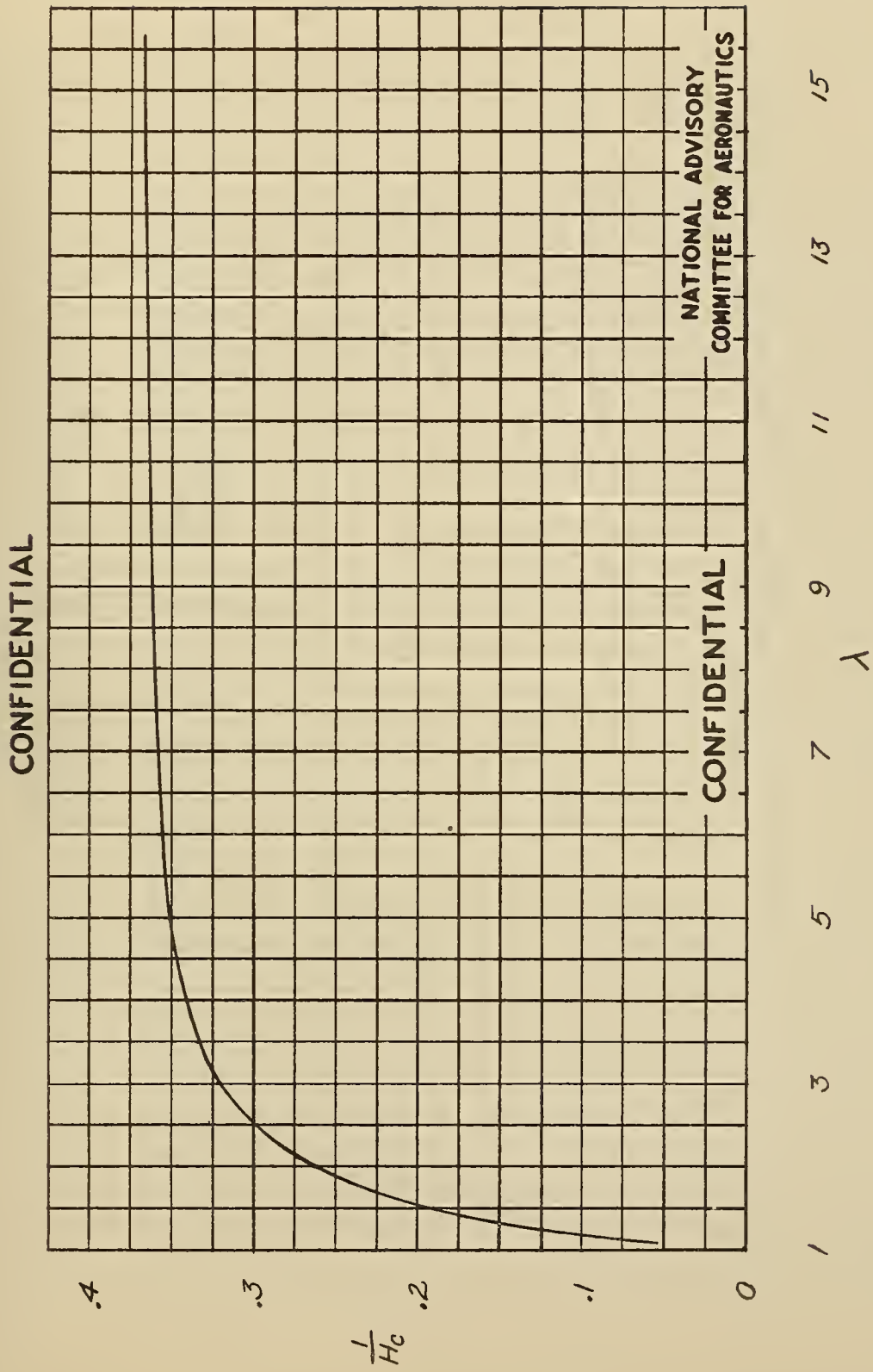
Figure 3.- Variation of $1/H_c$ with λ for the Blasius flat-plate velocity distribution.

Fig. 4   NACA ACR No. L6A22
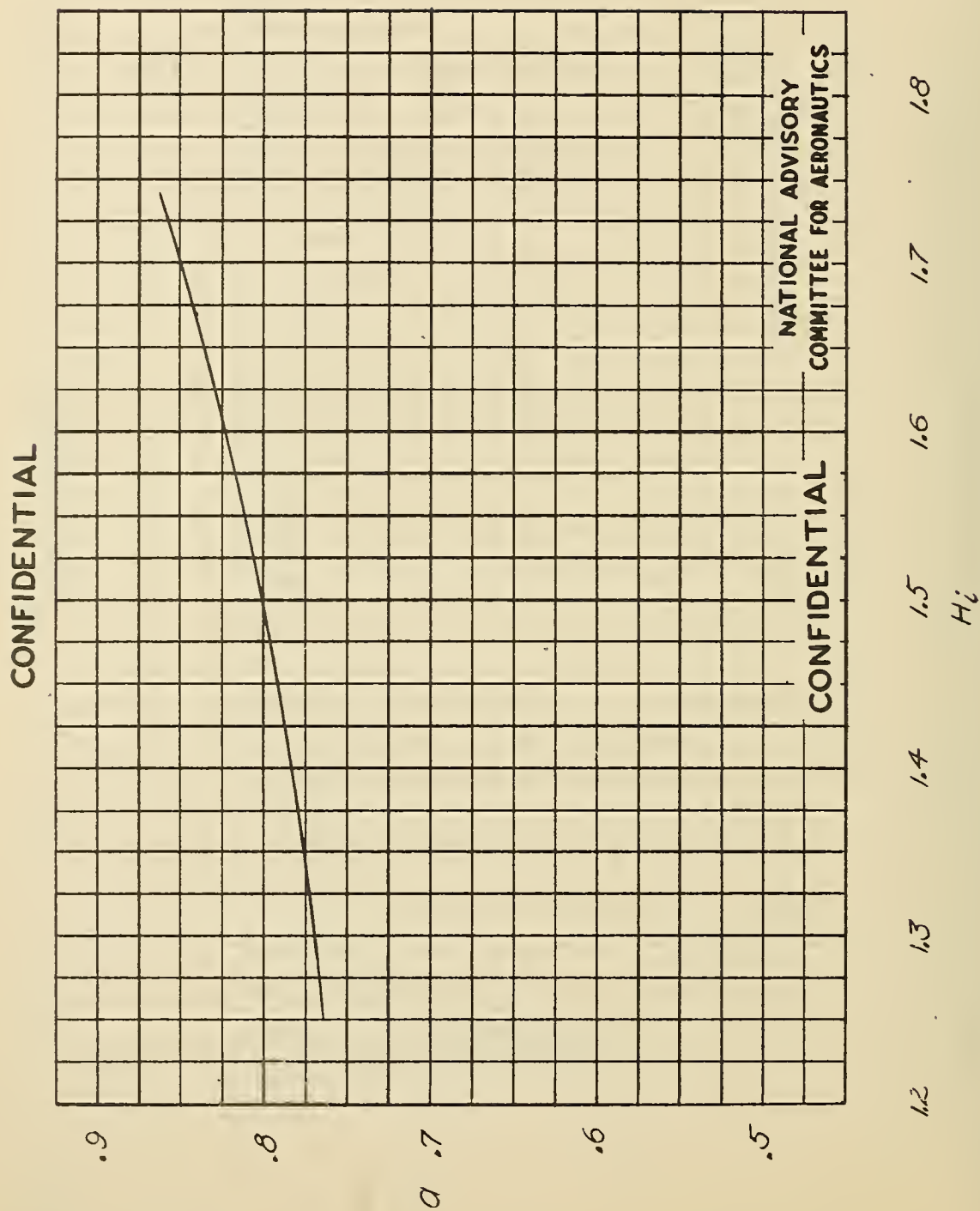
Figure 4.- Variation of a with H_i.

Figure 5.- Variation of θ/δ with λ for various values of H_1.

Fig. 6
NACA ACR No. L6A22
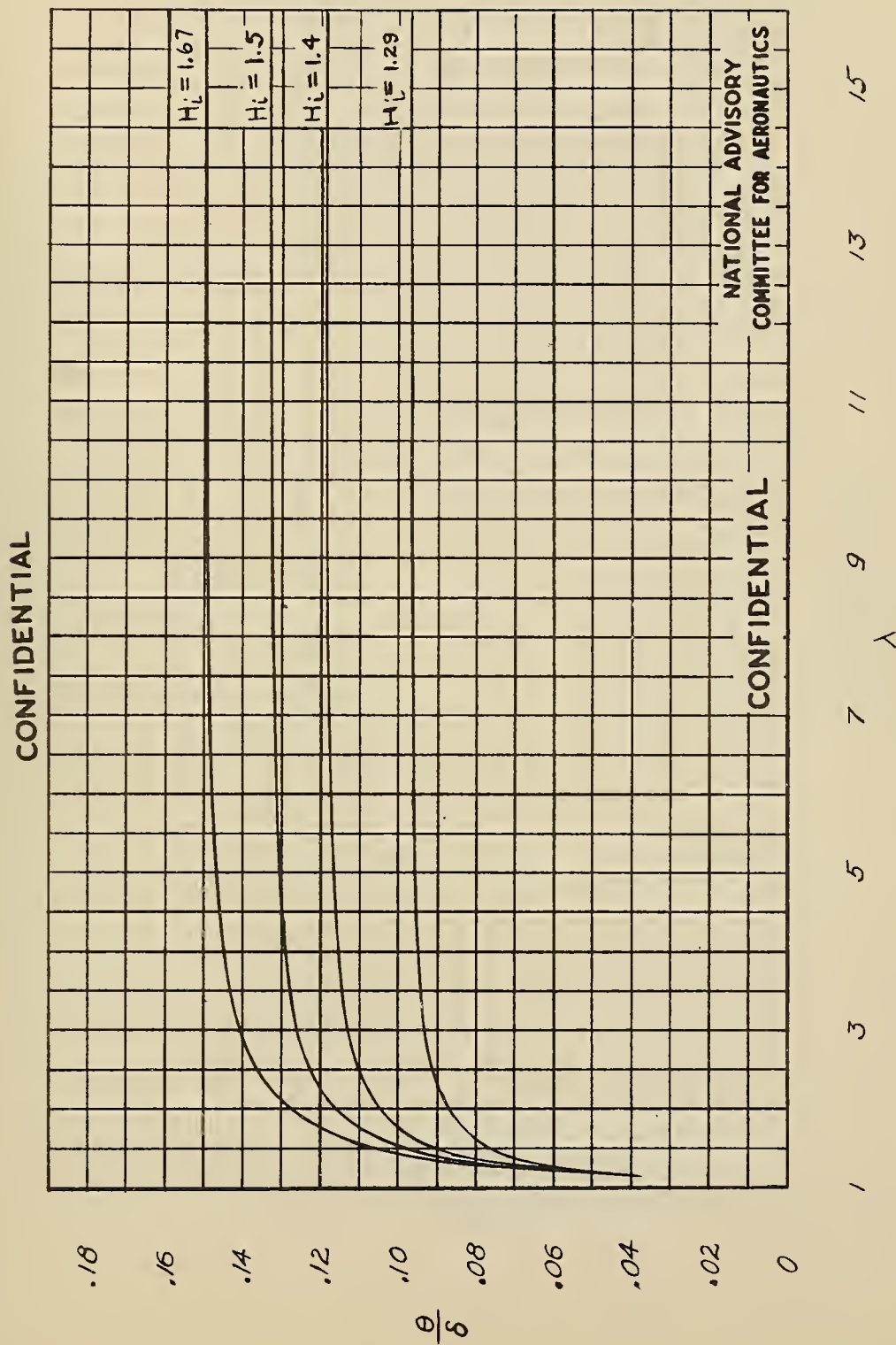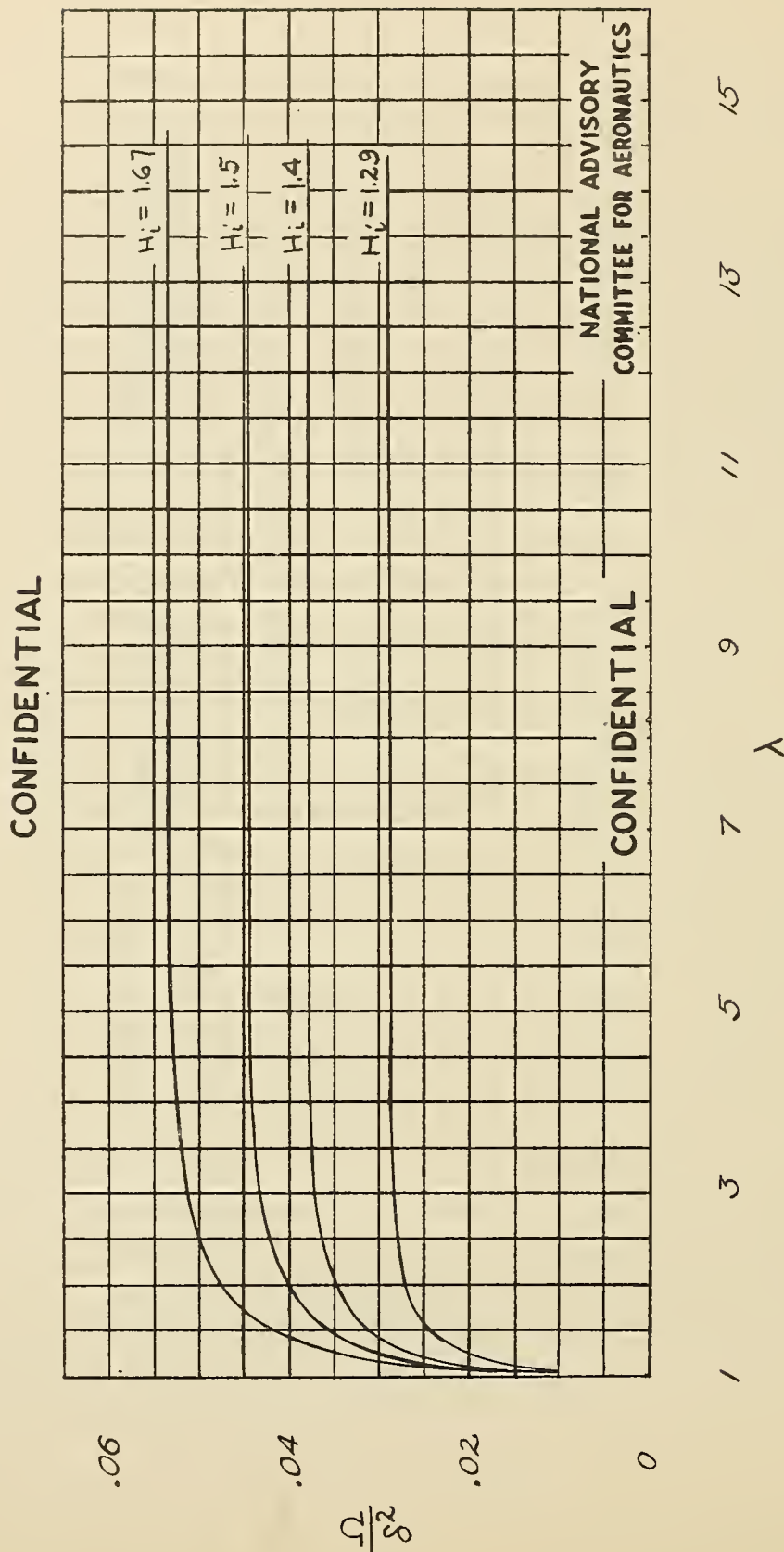
Figure 6.- Variation of Ω/δ^2 with λ for various values of H_i.
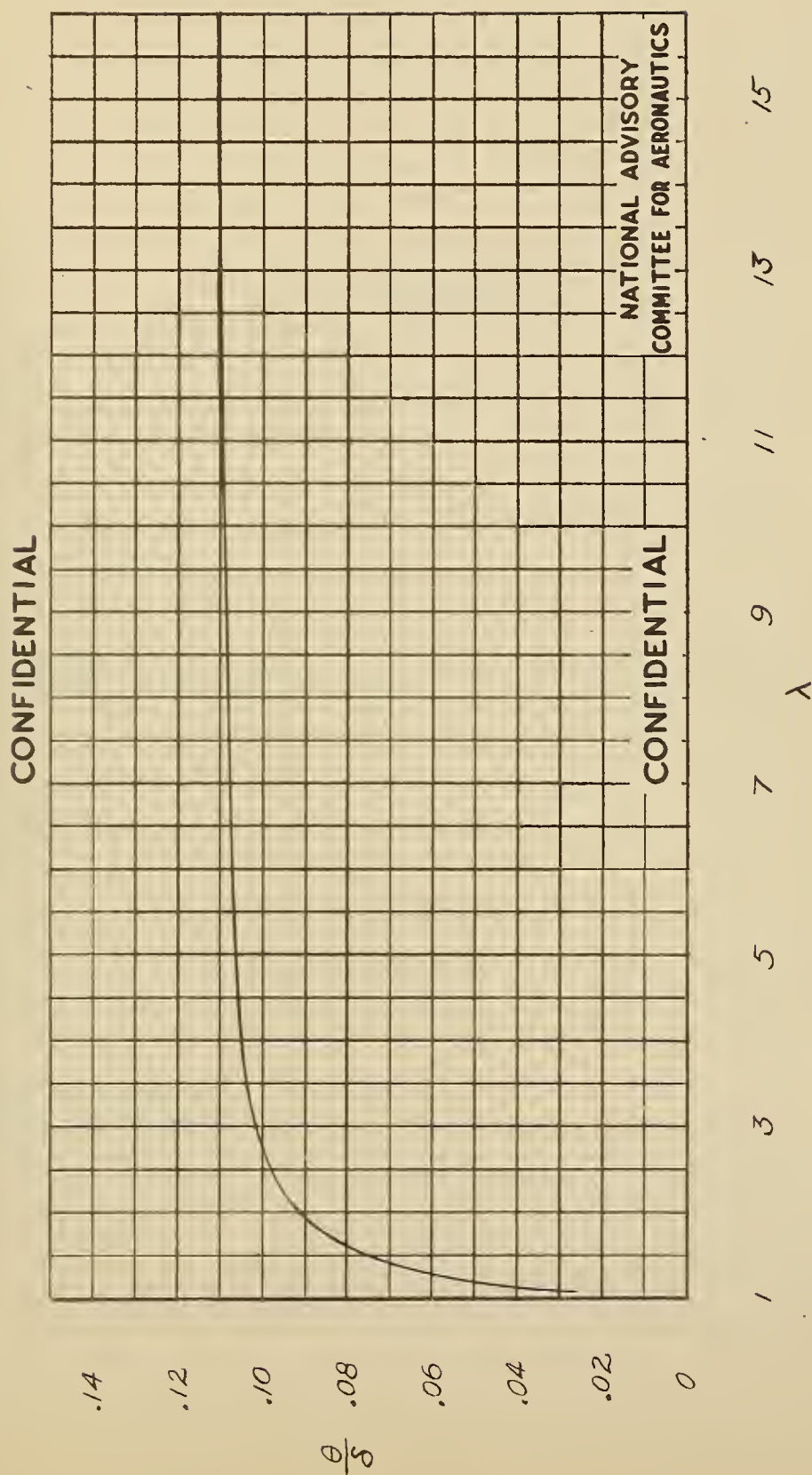
Figure 7.— Variation of θ/δ for the Blasius flat-plate profile with λ.

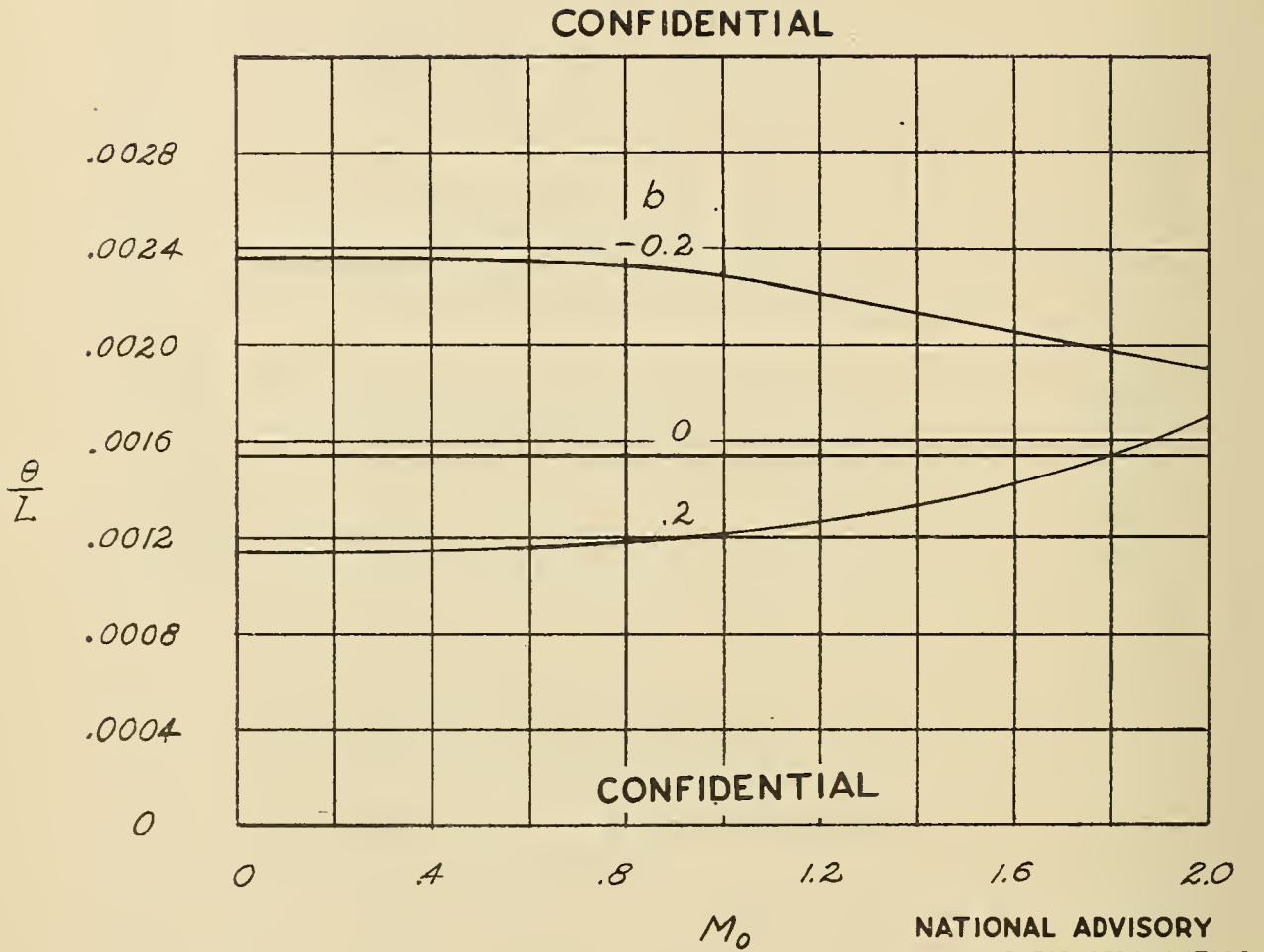Fig. 8   NACA ACR No. L6A22

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Figure 8.- Variation of θ/L with M_0 for velocity distribution given by $\dfrac{U}{U_0} = 1 + b\dfrac{x}{L}$; $0 < x/L < 1$. Values of constants in equation (14): $k = 0.006535$; $n = 1/6$; $R_L = 10^7$; $w = 0.768$; $\gamma = 1.4$; $H_1 = 1.4$; $a = 0.78$; $\dfrac{\theta}{L}' = 0$.